ABSTRACT: Flow field around rotors in axial flight is known to be complex especially in steep descent where the rotor is operating inside its own wake. It is often reported that, in this flight condition, the rotor is susceptible to severe wake interactions causing unsteady blade load, severe vibration, loss of performance, as well as poor control and handling. So far, there is little data from experimental and numerical analysis available for rotors in axial flight. In this paper, the steady Reynolds-Averaged Navier-Stokes Computational Fluid Dynamics solver Helicopter Multi-Block was used to predict the performance of rotors in axial flight. The main objective of this study was to improve the basic knowledge about the subject and to validate the flow solver used. The results obtained are presented in the form of surface pressure, rotor performance parameters, and vortex wake trajectories. The detailed velocity field of the tip vortex for a rotor in hover was also investigated, and a strong self-similarity of the swirl velocity profile was found. The predicted results obtained when compared with available experimental data showed a reasonably agreement for hover and descent rate, suggesting unsteady solution for rotors in vortex-ring state.

KEYWORDS: Axial flight, CFD, Performance, Rotor, Wake structure.

INTRODUCTION

In axial flight, particularly in steep descent where helicopter enters vortex-ring state (VRS), a rotor descends into its own wake. VRS is one of the extreme axial flight conditions and a potentially hazardous flight regime of operation where rotor blades are known to experience severe thrust fluctuations, highly unsteady blade airloads, aperiodic blade flapping, severe vibration, and sudden loss of aircraft altitude and control (Drees and Hendal 1951; Brinson and Ellenrieder 1998; Conlisk 2001; Leishman 2006; Brand et al. 2011). This problem is even more dangerous when helicopters operate close to the ground, which may cause unwanted accidents.

Several experiments and computational studies for low speed descending flight have been reported in literature, but relatively a few show detailed quantitative measurements of rotor performance and blade aerodynamic characteristics. Drees and Hendal (1951) are among the first to capture the flow field around rotors in axial flight. The airflow near a rotor installed in the wind tunnel was made visible by using a number of smoke generators. From the experiment, a large flow circulation was captured at the tip of the blade in VRS, and a periodic tumbling motion of the rotor was observed. Another study on measuring rotor performance in axial flight was carried out by Caradonna (1999). The climb performance of a scale model of UH-1H rotor was measured in the wind tunnel at a constant blade tip speed combined with various climb rates and blade collective pitch angles. The airflow near a rotor installed in the wind tunnel was made visible by using a number of smoke generators. From the experiment, a large flow circulation was captured at the tip of the blade in VRS, and a periodic tumbling motion of the rotor was observed. Another study on measuring rotor performance in axial flight was carried out by Caradonna (1999). The climb performance of a scale model of UH-1H rotor was measured in the wind tunnel at a constant blade tip speed combined with various climb rates and blade collective pitch angles. The airflow near a rotor installed in the wind tunnel was made visible by using a number of smoke generators. From the experiment, a large flow circulation was captured at the tip of the blade in VRS, and a periodic tumbling motion of the rotor was observed.
also compared to the HELIX-I code. The experimental results provided good quantitative rotor performance and a wake geometry database for Computational Fluid Dynamics (CFD) validation. The difficulties to get accurate rotor performance in hover from experiments, however, required an extrapolation technique to the limit of 0 climb speed. Instead of using a wind tunnel, Washizu et al. (1966) used a towing basin to measure a single rotor performance in axial flight. The use of large towing basin in their research was intended to eliminate blockage effects encountered in wind tunnels with closed test sections. Rotor performance for a range of descent speeds was presented in the form of a polar curve and compared with momentum. There are several other experiments on scaled rotor in hover from experiments, however, required an extrapolation to validate the flow solver. The results obtained are presented as surface pressure, rotor performance parameters, and vortex wake trajectories. The detailed velocity field of the tip vortex for rotor in hover is also investigated, and a strong self-similarity of the swirl velocity profile is found. The predicted results obtained were compared with available experimental data and showed a reasonably agreement for hovering and moderate rate of descent.

**CFD FLOW SOLVER**

The description of the HMB is discussed in detail by Barakos et al. (2005) and Steijl et al. (2006), being briefly presented in this paper. HMB is a computational fluid dynamic Navier-Stokes solver developed at the CFD Laboratory of the University of Liverpool and runs on parallel distributed memory computers. HMB solves the 3D Unsteady Reynolds-Averaged Navier-Stokes (URANS) equations on multi-block structured meshes using a cell-centred finite-volume method for spatial discretisation. In this study, the convective terms are discretised using a k-ω model for the outer boundary layer. Furthermore, the redefined SST eddy viscosity accounts for the effect of transport of turbulent shear stress. The SST model has been used for many external flows with adverse pressure gradients. The solver also has been used and validated for several fundamental flows apart from rotor cases (Badcock et al. 2000; Gaggiardi and Barakos 2008).

Hovering rotor flow simulations are performed by assuming that the wake shed from the rotor is steady. The hover formulation of HMB allows the computation of hovering rotor flows to be treated as steady-state cases. The formulation uses a mesh that does not rotate and employs a transformation of the frame of reference to account for the rotation. Applying periodic boundaries, an n-bladed rotor can be approximated using a 1/n domain segment. The details of the hover formulation
can be found in Steijl et al. (2006). For hovering and climbing rotor cases, the momentum based far field boundary model, assuming a three-dimension source-sink singularity at the rotor axis of rotation and at a rotor plane was used (Steijl et al. 2006; Srinivasan 1993). For ascending and descending rotor, the vertical velocity is added to the hover velocity. The Froude/Source-sink condition typically used (Steijl et al. 2006) for applying the boundary conditions in hover and descending flight at moderate speeds was not adequate for the extreme cases considered here. Therefore, in this paper, for extreme case in descending flight, a free-stream condition was assumed at the far field. For more efficient computations, the vortex-tube model that valid for most of axial flight condition can be used (Wang 1990; Perry et al. 2007; Nik Mohd and Barakos 2010).

**CFD MESH GENERATION AND TEST CASES**

A structured C-H-H CFD mesh was employed (Fig. 1). An H-type structure was used away from the blades with a C-type structure attached to them. This type of mesh topology allows for accurate viscous computations and can provide a mechanism for pitching the blades, with the near-blade mesh remaining in an undeformed state (Brocklehurst et al. 2008). For all meshes, viscous spacing in the direction normal to the blade surface with the first cell located at $10^{-5}c$ above the blade and exponential distribution was employed. For viscous cases, this mesh spacing constructed results in $y^+ < 1$. The rotor centre of rotation is located at the z-axis, the rotor blade is laid on the x-axis, and the quarter chord point is taken at $y = 0$ with the blade leading edge pointing at the positive y-axis. This geometry was used for the construction of the computational meshes for the rotors in axial flight. The rotor hub was modelled as a simple straight cylinder. Summary of rotor geometry and flow condition considered in this study are shown in Table 1.

**RESULTS AND DISCUSSION**

**ASCENDING CARADONNA UH-1H ROTOR**

For ascending case, the aerodynamic performance and wake geometry of the Caradonna UH-1H rotor were computed for a range of ascend rates, $V_v/V_{tip}$, from hover to ascending rate of 0.04. The results are presented for a blade at trimmed and untrimmed conditions. For untrimmed condition, the rotor blade collective pitch angle, $\theta_o$, was set at 11° as in Caradonna (1999) experiment. For trimmed conditions, the thrust required for ascend was maintained constant closed to the hover state (Fig. 2a). This is done by considering the incremental of the blade collective pitch settings by $\frac{3}{4} \frac{V_v}{V_{tip}}$ (Johnson 1980). The coning angle was set at 0 for all calculations. For the ascending test cases considered, the linear increments in the blade collective pitch resulting from the trimmed conditions are plotted in Fig. 2b. The blade tip Mach number was $M_{tip} = 0.5771$ and the Reynolds number, $Re = 1.028 \times 10^6$.

Rotor trimming requires an increment in blade collective pitch angle to maintain the rotor thrust as the ascend speed ratio is increased. This also results in an increment of the rotor torque required (Fig. 2c). For the test cases considered, the rotor losses its efficiency in ascending flight. Comparison with the experimental measurements (Fig. 2d) shows that HMB underpredicts rotor figure of merit (FM) but is in good agreement with HELIX-I data for ascending cases.

**Table 1. Summary of rotor geometries for axial flight test cases.**

<table>
<thead>
<tr>
<th>Characteristic</th>
<th>UH-1H</th>
<th>Washizu</th>
</tr>
</thead>
<tbody>
<tr>
<td><strong>Rotor geometry</strong></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Number of blades</td>
<td>2</td>
<td>3</td>
</tr>
<tr>
<td>Root cut-out</td>
<td>1 c</td>
<td>1 c</td>
</tr>
<tr>
<td>Rotor radius</td>
<td>13.67 c</td>
<td>16.67 c (1.1 m)</td>
</tr>
<tr>
<td>Chord</td>
<td>1 c</td>
<td>1 c (0.33 m)</td>
</tr>
<tr>
<td>Aspect ratio</td>
<td>13.67</td>
<td>16.67</td>
</tr>
<tr>
<td>Twist</td>
<td>Untwisted</td>
<td>−8.35°</td>
</tr>
<tr>
<td>Solidity</td>
<td>0.047</td>
<td>0.0573</td>
</tr>
<tr>
<td>Blade profile</td>
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<td>NACA 0012</td>
</tr>
<tr>
<td><strong>Flow condition</strong></td>
<td></td>
<td></td>
</tr>
<tr>
<td>Tip Mach</td>
<td>0.5771</td>
<td>0.1693</td>
</tr>
<tr>
<td>Reynolds number ($\times 10^5$)</td>
<td>1.028</td>
<td>0.1306</td>
</tr>
<tr>
<td>Grid size ($\times 10^6$)</td>
<td>9</td>
<td>6.3</td>
</tr>
<tr>
<td>Number of blocks</td>
<td>170</td>
<td>272</td>
</tr>
<tr>
<td>Test case</td>
<td>Hover/climb</td>
<td>Hover/descent</td>
</tr>
</tbody>
</table>

**Figure 1.** Multi-block mesh employed in HMB.
Predictions of hover FM were successfully presented in Nik Mohd et al. (2011) for the case of Caradonna and Tung (1981). There are, however, some discrepancies for the UH-1H rotor for both HMB and HELIX-I data. In experiments, the drop in the value of FM is expected due to excessive flow unsteadiness (Caradonna 1999), and extrapolation to ascend was suggested to correct the hover FM. The predicted hover FM shows a linear trend; however, the value of the FM falls between the experiments and the HELIX-I data. Overall, for the ascending rotor, the predicted rotor performance without trimming is in excellent agreement with experiments and HELIX-I data. Nevertheless, when the rotor trim condition is activated, a slight increase is observed in the rotor FM for the fast ascending case (Fig. 2d).

Figures 3a to 3d show the blade surface pressure coefficient, $C_P$, at 3 blade stations plotted for different ascend rates. From the comparison, the blade surface pressure is found to have the same stagnation values, but decreasing suction pressure, as rotor ascend rate increases. The distributions of spanwise blade lift

![Figure 2](image_url)

**Figure 2.** Performance of the Caradonna UH-1H rotor in hover and vertical ascent. (a) Thrust coefficient; (b) Collective pitch; (c) Torque coefficient; (d) Figure of merit.

![Figure 3](image_url)

**Figure 3.** Comparison of CFD results of the Caradonna UH-1H blade surface pressure coefficients at different ascent rates. (a) $r/R = 0.30$; (b) $r/R = 0.70$; (c) $r/R = 0.90$; (d) $r/R = 0.99$; (e) Lift coefficient. ROC: Rate of climb.
Performance and Wake Analysis of Rotors in Axial Flight Using Computational Fluid Dynamics

The distributions of vortex swirl velocity normalised with the rotor tip speed across the tip vortex cores for various wake ages are shown in Fig. 6. This information also shows how strong the axial velocity induced by the tip vortex is. The vortex swirl velocity of the tip vortex weakens as it evolves

Figure 5. Superimposed vorticity cuts at Ψw = O°.

Figure 6. Distribution of vortex swirl velocity of the UH-1H rotor in ascending flight. (a) Vv/Vtip = 0 (Hover); (b) Vv/Vtip = 0.04.
but grows in core radius. Furthermore, for the same wake age considered, the vortex swirl velocity is found to be stronger in hover compared to the ascending flight. This quantity may be influenced by the dissipation of vorticity due to coarse mesh constructed below the rotor plane where an exponential point distribution was used.

The growth of the tip vortex core extracted for one rotor revolution is presented in Fig. 7a. It is found that, for the same mesh distribution, the core radius increases linearly with vortex age and grows faster in hovering flight. Similarly to the hovering cases, the tip vortex of an ascending rotor looks self-similar when the normalised swirl velocity of the core is plotted against the normalised radial distance (Fig. 7b).

![Figure 7](image_url)

**Figure 7.** Predicted core growth and self-similarity of the UH-1H rotor tip vortices at different ascent rates. (a) Vortex core radius; (b) Vortex self-similarity.

**DESCENDING WASHIZU ROTOR**

Experiment for rotors in descending flight performed by Washizu et al. (1966) were referred in this study. In the experiment, the rotor was tested in a towing basin with the aim of eliminating the blockage effect that occurs in most wind tunnel experiments.

The experiments of the Washizu rotor in descending flight include hover, vortex-ring state \((V_{v}/V_{tip} = 0.054, V_{v} = 3.11 \text{ m/s})\) and wind-mill brake state, WBS \((V_{v}/V_{tip} = 0.09, V_{v} = 5.184 \text{ m/s})\), and these were computed using CFD. The test cases consider the blade tip Reynolds and Mach numbers fixed to a constant value of \(1.306 \times 10^6\) and 0.1693, respectively. The rotor has 3 blades, and the blade collective pitch setting varied from 0°, 4.5° to 8°. The descending flight was simulated by increasing the descend speed while keeping the rotor tip speed constant. In CFD, the freestream far field boundary was specified for rotors in descending flight, and a Froude (source-sink) condition was used for the hovering flight case. The turbulence closure was achieved using the 2 equations of the \(k-\omega\) model with transport of shear stress (SST) and baseline (BSL) variant. Figures 8a and 8b show the convergence history for rotor thrust and torque coefficients. It can be seen that rotor thrust and torque in WBS fluctuate even after 80,000 iteration steps. This behaviour is apparently caused by the highly separated flow developed on the upper blade surface. In descending flight, the rotor blades experience a flow coming from below the rotor plane.

![Figure 8](image_url)

**Figure 8.** Convergence history plots for the Washizu rotor in descending flight \((\theta = 8°, M_{tip} = 0.1693)\). (a) Thrust history; (b) Torque history.
Figs. 10 and 11. As shown in Fig. 10, for the rotor in VRS, the blade at 20%R experiences leading edge separation and then growth to fully separated flow on the upper blade surface at 40%R. A much more severe flow separation can be verified for the blade in windmill brake as shown in Fig. 11, where almost 75%R of blade is separated.

For hovering rotor, the flow field enters the computational domain from all directions and is induced downstream below the rotor. In descending flight, however, the inflow velocity enters the domain from the boundary located below the rotor plane. In the VRS, where the transport of vorticity is approximately equal to the rate of descent, a large recirculation...
flow exists near the blade tip. Nevertheless, the mesh setting used was still unable to accurately capture the collection of tip vortices accumulated in this recirculation flow as found in most experiments. In the windmill brake state, where the rotor experiences a fast descent rate, the flow field continues to expand above the rotor plane. Iso-surfaces of Q-criterion coloured by pressure are depicted in Fig. 12. The Q-criterion is used here to visualise the highly concentrated vorticity regions by plotting the rotation-dominated strain in the flow. A large area of fully turbulence flow is seen to be developed around the rotor in WBS.

The comparisons of the rotor performance in terms of torque coefficient ($C_Q$) versus thrust coefficient ($C_T$) and FM versus $C_T$ are given in Fig. 13. The effects of different turbulence models ($k-\omega$ BSL and SST) and trim conditions are also presented. The results suggest that, for low collective pitch settings, the CFD slightly overpredicts rotor torque for all test cases. Trimming the rotor improves the performance prediction compared to the untrimmed condition. Moreover, for a high collective pitch setting, a large offset of rotor torque was observed.

The FM of rotors in hover is plotted for different blade collective pitch settings in Fig. 13b. The CFD results show a well predicted FM for non-lifting rotor ($C_T = 0$), but this value is slightly below the experimental results for the 4.5° and 8° pitch settings. From these results, it can be concluded that the CFD method used can predict rotor performance in steep descent up to an acceptable accuracy.

Comparisons of the predicted rotor performance (trimmed condition) against momentum theory and experiments are shown in Fig. 14. As explained in the previous section, the rotor was trimmed to a specific thrust coefficient. The effect of trimming can be seen on the torque coefficient required to maintain the rotor descend speed for a particular rotor thrust and blade collective angles. There are discrepancies in torque prediction between CFD and the other 2 data. These could be due to potential errors in the experiments at high pitch angles and high rates of descent due to limited speed of the carriage, and the length of the track (Washizu et al. 1966).
Furthermore, no detailed discussion on the hover and windmill test cases was given.

\[ \text{Figure 14.} \text{ Comparison of the Washizu rotor performance plotted for different descent rates and collective pitch settings.} \]

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**CONCLUSIONS**

Accurate prediction of rotor blade aerodynamics in axial flight is a challenging task in computational fluid dynamics. This is due to the presence of large-scale vortical structures that interact with the blade near the tip and need to be accurately modelled. In this study, the performance of rotors in various axial flight conditions in vertical ascend, hover, and descent was computed using the Helicopter Multi-block (HMB) flow solver.

For rotors in hover and vertical ascend, the blade surface pressure, the integrated rotor performance, and the vortex wake trajectory were analysed using HMB and compared with the experimental data of the UH-1H rotor. It can be concluded that the HMB solver can predict well the blade aerodynamic performance in comparison to experimental and HELIX-I data. However, for low ascending rate, small discrepancies can be observed. This is expected due to the aperiodicity and recirculation of the flow field in the wind tunnel rather than CFD errors. Similar hover test cases were also employed for validation including the hovering Caradonna-Tung model rotor, with good success. For the same wake age considered, the distribution of the normalised vortex swirl velocity across the vortex core in climb was found to be weaker and smaller in radius than for the rotors in hover. Furthermore, the tip vortex showed more contraction in hover and fast descending than for ascending flight.

When the rotor descended into its own wake, a large flow separation developed on the upper surface of the blade extending up to 55%R. A more severe and chaotic flow field was observed for the rotor in windmill brake state where more than 75%R of the blade span experienced stall. The predictions of thrust and torque showed that the blade collective pitch had to be trimmed to the required thrust. The used of the baseline (BSL) and shear stress transport (SST) k-ω turbulence models showed little influence on the torque predictions.

Overall, the present work on rotors in axial flight has successfully validated the HMB solver using several rotor test data. The work presented, provides additional knowledge on rotor performance but there are still certain areas that are not well understood. Although CFD can be a good tool in predicting the basic flow physics on different axial flight regimes, future work and validation against new experiments are needed. This calls for experiments in various regimes of axial flight that should be the target of large international initiatives and should result in open databases of measurements.

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**AUTHOR’S CONTRIBUTION**

Mohd NARN performed the numerical calculation and prepared the figures; Barakos G conceived the idea and co-wrote the main text. All authors discussed the results and commented on the manuscript.
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